AD NUMBER
AD035545
CLASSIFICATION CHANGES
TO: **UNCLASSIFIED**
FROM: **CONFIDENTIAL**
LIMITATION CHANGES
TO: **Approved for public release; distribution is unlimited.**
FROM: **Distribution authorized to U.S. Gov't. agencies and their contractors; Administrative/Operational Use; 15 JUN 1954. Other requests shall be referred to Naval Proging Ground, Dahlgren, VA.**
AUTHORITY
30 Jun 1966, DoDD 5200.10; NWL notice dtd 7 Sep 1972

AD _____

CONFIDENTIAL

NOTICE: THIS DOCUMENT CONTAINS INFORMATION AFFECTING THE NATIONAL DEFENSE OF THE UNITED STATES WITHIN THE MEANING OF THE ESPIONAGE LAWS, TITLE 18, U.S.C., SECTIONS 793 and 794, THE TRANSMISSION OR THE REVELATION OF ITS CONTENTS IN ANY MANNER TO AN UNAUTHORIZED PERSON IS PROHIBITED BY LAW.

CONFIDENTIAL

NPG Report No. 1279

IGNITION OF LIQUID PROPELLANT IN THE 40MM GUN

U. S. NAVAL PROVING GROUND
DAHLGREN, VIRGINIA

Copy No. 35

CONFIDENTIAL

Date : 15 June 1954

CONTENTS

ABSTRACT

The tests reported here were conducted to determine the effects produced on the ballistics of a liquid monopropellant round by advancing the point of venting of the primer within the propellant column. The tests were conducted with two case configurations in which the L/D ratios of the propellant columns were approximately 3.4 and 5.0. As the point of venting was advanced, muzzle velocity increased in rounds with the L/D ratio of 5.0, but decreased in those with the L/D ratio of 3.4. The amount of unburned propellant remaining in the case after firing increased in both case configurations. In both configurations, the pressure curves were transformed from curves with two distinct pressure peaks, through stages approximating plateau configurations, to curves with one well defined pressure maximum.

FOREWORD

This is the Second Partial Report on Task Assignment NPG-Re5a-39-1-53, "Liquid Propellant Guns". Reference (a) established the task and authorized the use of funds under this task for the development of a liquid propellant round and the study of erosion characteristics of liquid propellants in guns. This is also the Second Partial Report on Task Assignment NPG-Re2d-12-1-53, "Liquid Propellant for Guns; test and evaluation of". The object of this task, established by reference (b), was to develop a prepackaged hydrazine monopropellant round and test its performance in automatic fire. Since both objectives require the development of a satisfactory round before either rapid fire or erosion tests can be conducted, the objectives of the two task assignments are combined in these tests. The experimental work described in this report was accomplished between 18 August 1953 and 14 September 1953.

This report was reviewed by:

S. E. HEDDEN, Acting Head
Interior Ballistics Division and Head
Research and Development Branch
Armament Department
H. S. OVERMAN, Director of Research
Armament Department
L. C. KLINGAMAN, Commander, USN
Armament Officer
Armament Department
N. A. M. RIFFOLT, Director of Research

INTRODUCTION

As a result of the research reported in reference (c) a hydrazine monopropellant round was developed by which service velocity in the 40mm gun could be exceeded by as much as 150 f/s without exceeding the chamber pressure of the solid propellant service charge for this gun. However, one characteristic of this round was the development of two pressure peaks with the second peak often equalling the first in magnitude. It was observed that the occurrence of this second maximum was frequently associated with more or less severe damage to the mouth of the case. The tests reported here were undertaken to determine if and to what extent the ballistics of the above round might be improved by relocating the point at which the primer vented in the propellant column. In particular, it was of interest to reduce the magnitude of the second pressure peak.

DESCRIPTION OF MATERIAL

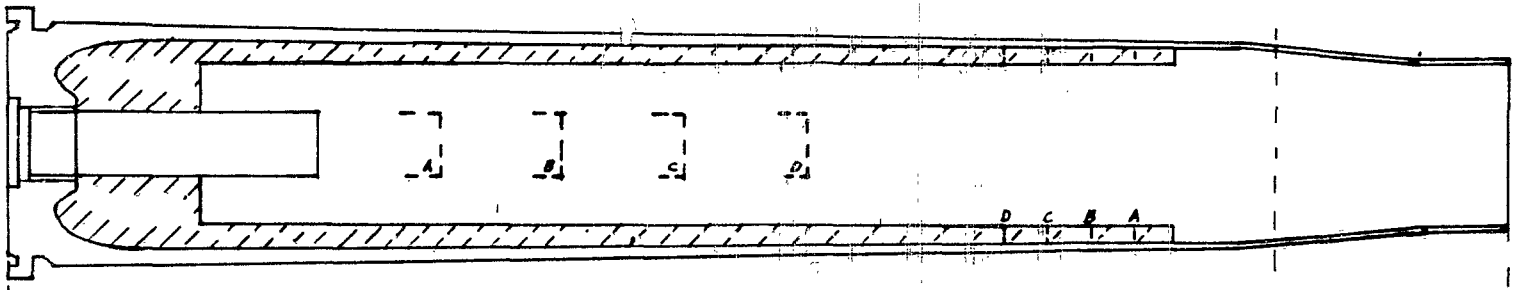
All the rounds used in these tests were loaded with 310 grams of hydrazine monopropellant of the following composition:

Hydrazine	73.5%
Hydrazine Nitrate	22.6%
Water	3.9%

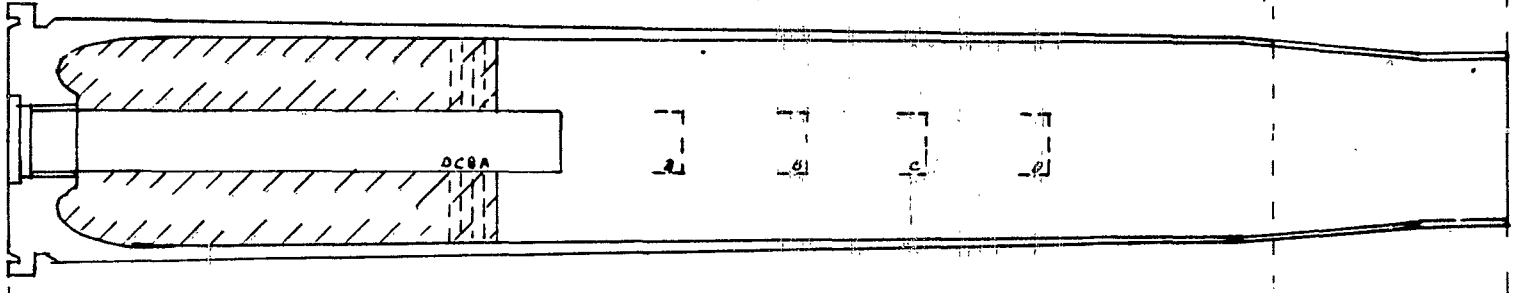
The two case assemblies, designated C-4 and C-5, produced L/D ratios of the propellant columns of approximately 5.0 and 3.4, respectively. A free volume of 1% was provided in the C-4 assembly and 5% in the C-5. The excess case volume was occupied by a paraffin beeswax mixture. This wax was distributed in the cases as shown in Figure 1 in which the wax is the hatched areas.

Cases were 40mm Mk 2 modified to receive Mk 42 primers.

ASSEMBLY C-4



ASSEMBLY C-5



Projectile Base

FIGURE NO. 1

SKETCH OF CASE ASSEMBLIES

CONFIDENTIAL

Primer tubes were cylindrical brass tubes with I.D.'s of 7/16 in. and 0.025 thick burst diaphragms at the ends of the tubes. The tubes were loaded with 0.2 grams of ammonium perchlorate and 36 grains of FFFG black powder. Lengths of tubes are shown in Figure 1 and tabulated in Tables 1 and 2.

Standard 40mm Mk 2 projectiles were used in all rounds.

DESCRIPTION OF TEST EQUIPMENT

The firings were conducted in a 40mm Mk A Mod 1 barrel mounted in a 6 pounder mount Mk 7 Mod 1. The barrel and the mount were modified to receive dynamic pressure gauges in the chamber 3.0 and 10.0 from the breech face. The instrumentation, including pressure gauges, used on these tests was described in reference (c).

PROCEDURE

The method of preparing and mixing the propellant was similar to that used in reference (c) except that temperatures up to 102°C and three or four days evacuation with the vacuum pump were required to expel water during the preparation of hydrazine nitrate. The adjustment of the case volume was also the same except that the filler was distributed in the case before the projectile was fitted and crimped.

The first series of firings comprised the rounds with the C-4 assembly. These were fired with primer extension tubes increasing in length from 3" to 6" by one inch increments. The volume occupied by the wax filler was reduced on succeeding rounds by the amount of increase in volume of the primer tubes as shown in Figure 1. Pressures were recorded in the gun chamber at 3" and 10" from the breech face.

The second series of firings comprised rounds with the C-5 assembly. A group of rounds with primer tubes 4" to 8" long were fired following the same procedure as above. A second group with tubes 4" to 7" in length was then fired as a reproducibility check.

RESULTS AND DISCUSSION

The effects produced by advancing the position of venting of the primer within the propellant column with the two different case assemblies may be seen by comparing the pressure time records reproduced in Figures 2 through 7, and by comparing the values for maximum pressure and muzzle velocities tabulated in Tables 1 and 2. The ejection times tabulated in the tables are the times from the occurrence of the first pressure spike to ejection of the projectile.

The C-4 Assembly

It is noted that, in general, the second pressure peak diminished as compared to the first pressure peak as the point of primer venting was moved forward in this assembly. The end of the tube was torn off on firing round 1 with the 3" primer, possibly because of the high secondary pressure registered by the 10"0 gauge. On the other hand, round 4 with the longest primer showed no secondary peak pressure on the oscillogram record. Velocities increased slightly with increased primer length except for round 3 which gave a value only slightly lower than the preceding round. The velocity increased from 2568 to 2675 f/s for the series. The velocities are somewhat lower than expected throughout the group even though freshly prepared propellant was used, but the primer length effects are, nevertheless, comparable.

TABLE 1 BALLISTIC DATA FROM C-4 ASSEMBLY

Rd. No.	Primer Length	Ejection Time (ms)	Pressure (psi) First Peak 3"0 Gauge	First Peak 10"0 Gauge	Second Peak 3"0 Gauge	Second Peak 10"0 Gauge	Velocity (f/s)
1	3"0	5.0	33060	32600	38100	41000	2568
2	4"0	4.7	36900	38100	31900	33850	2606
3	5"0	4.2	41650	39870	31175	32900	2594
4	6"0	4.0	51175	48850	---	---	2675

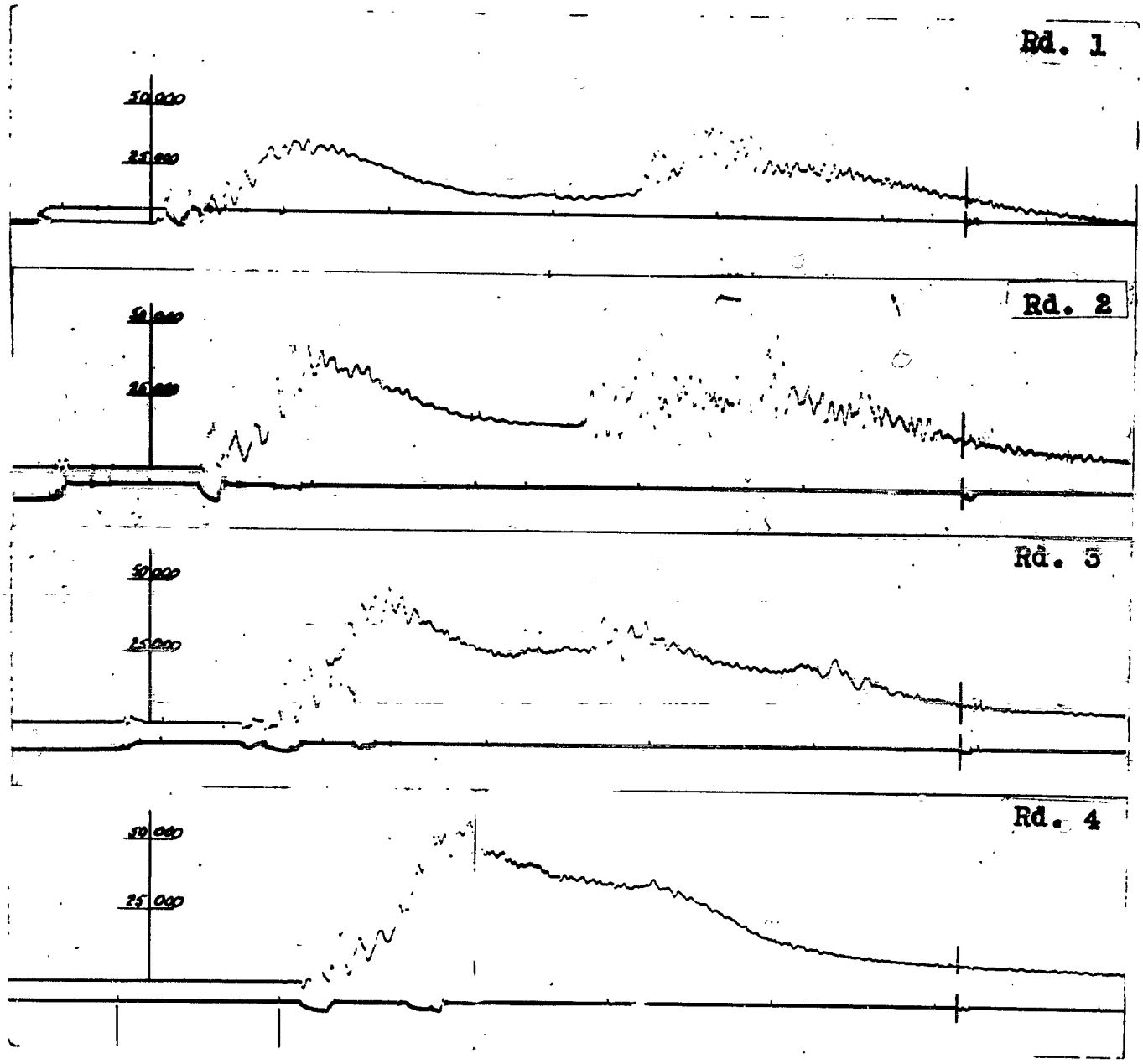


FIGURE NO. 2

PRESSURE-TIME OSCILLOGRAMS OBTAINED WITH ASSEMBLY C-4
AT THE 3" GAGE POSITION

CONFIDENTIAL

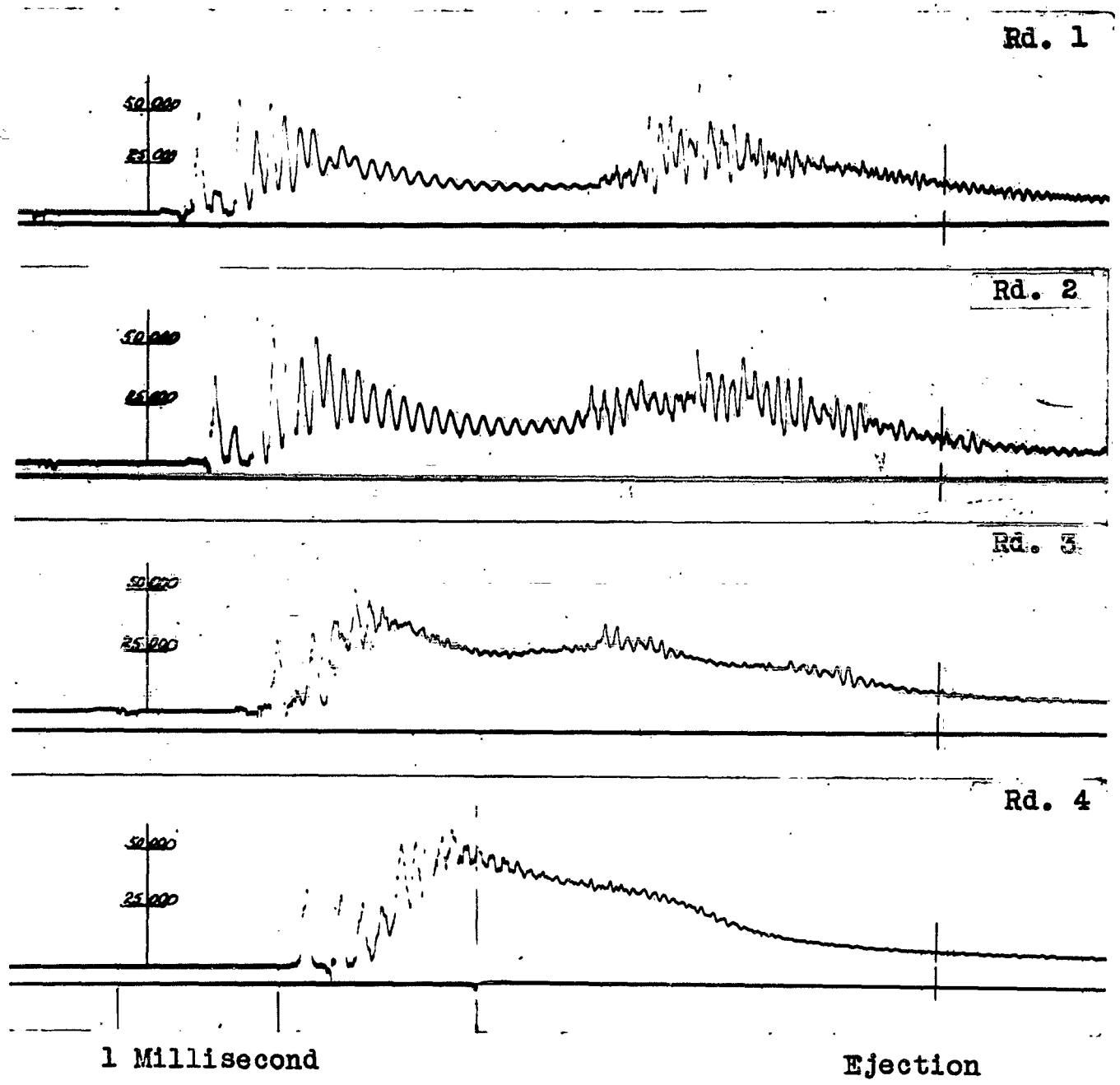


FIGURE NO. 3

PRESSURE-TIME OSCILLOGRAMS OBTAINED WITH CASE ASSEMBLY C-4
AT THE 10" GAGE POSITION

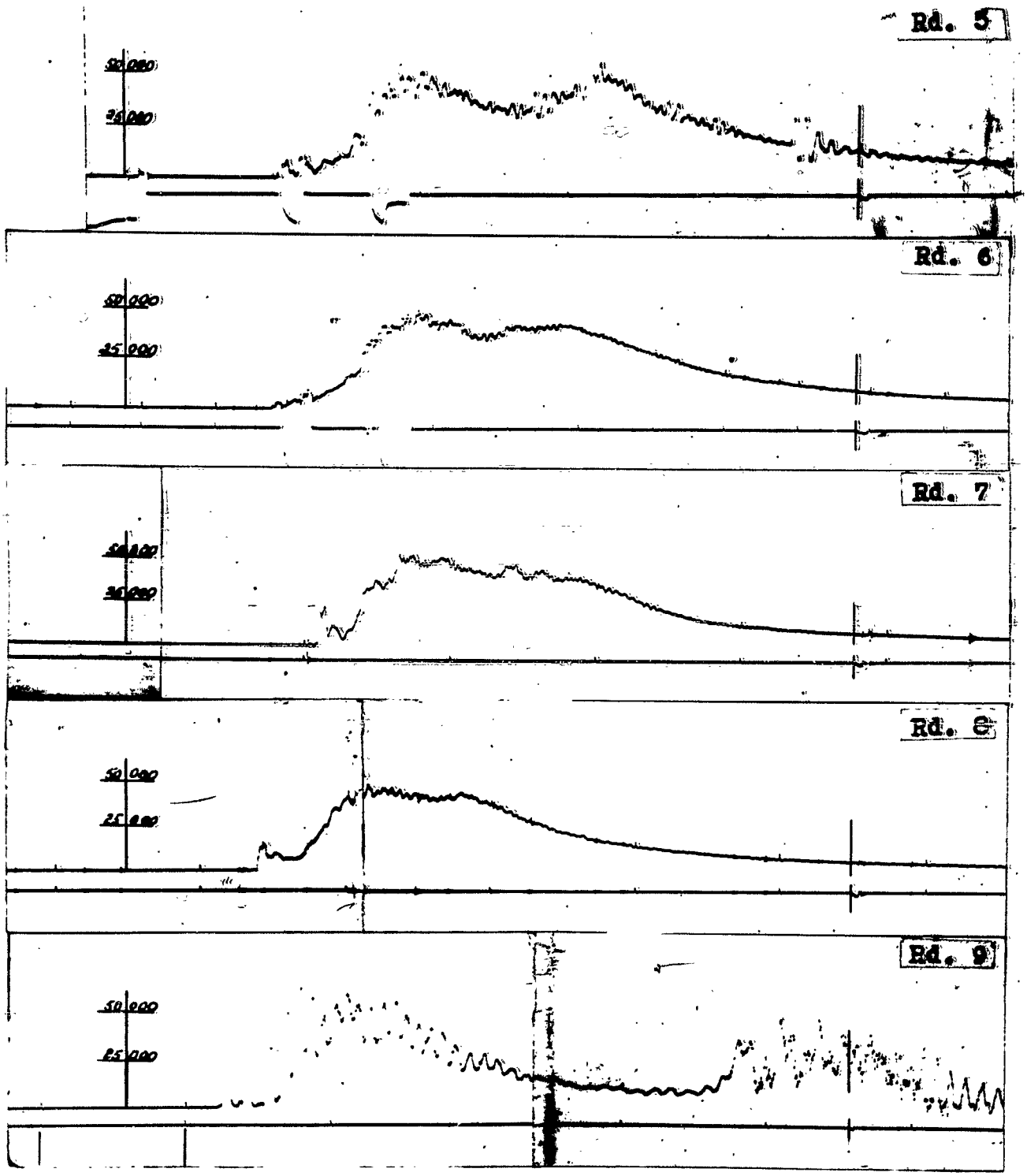


FIGURE NO. 4

PRESSURE-TIME OSCILLOGRAMS OBTAINED WITH CASE ASSEMBLY C-5 AT THE 3" GAGE POSITION

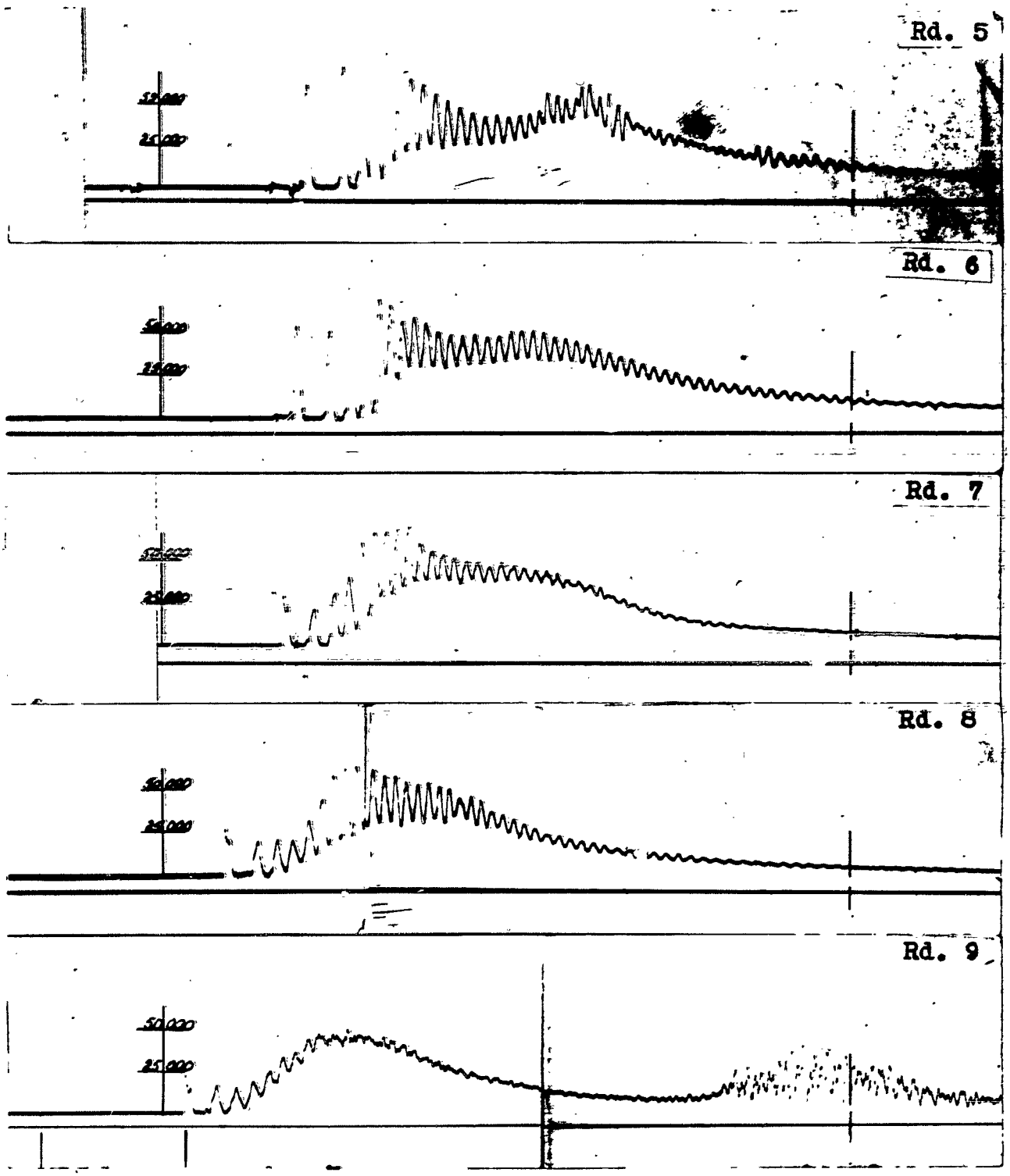


FIGURE NO. 5

PRESSURE-TIME OSCILLOGRAMS OBTAINED WITH CASE ASSEMBLY C-5 AT THE 10" GAGE POSITION

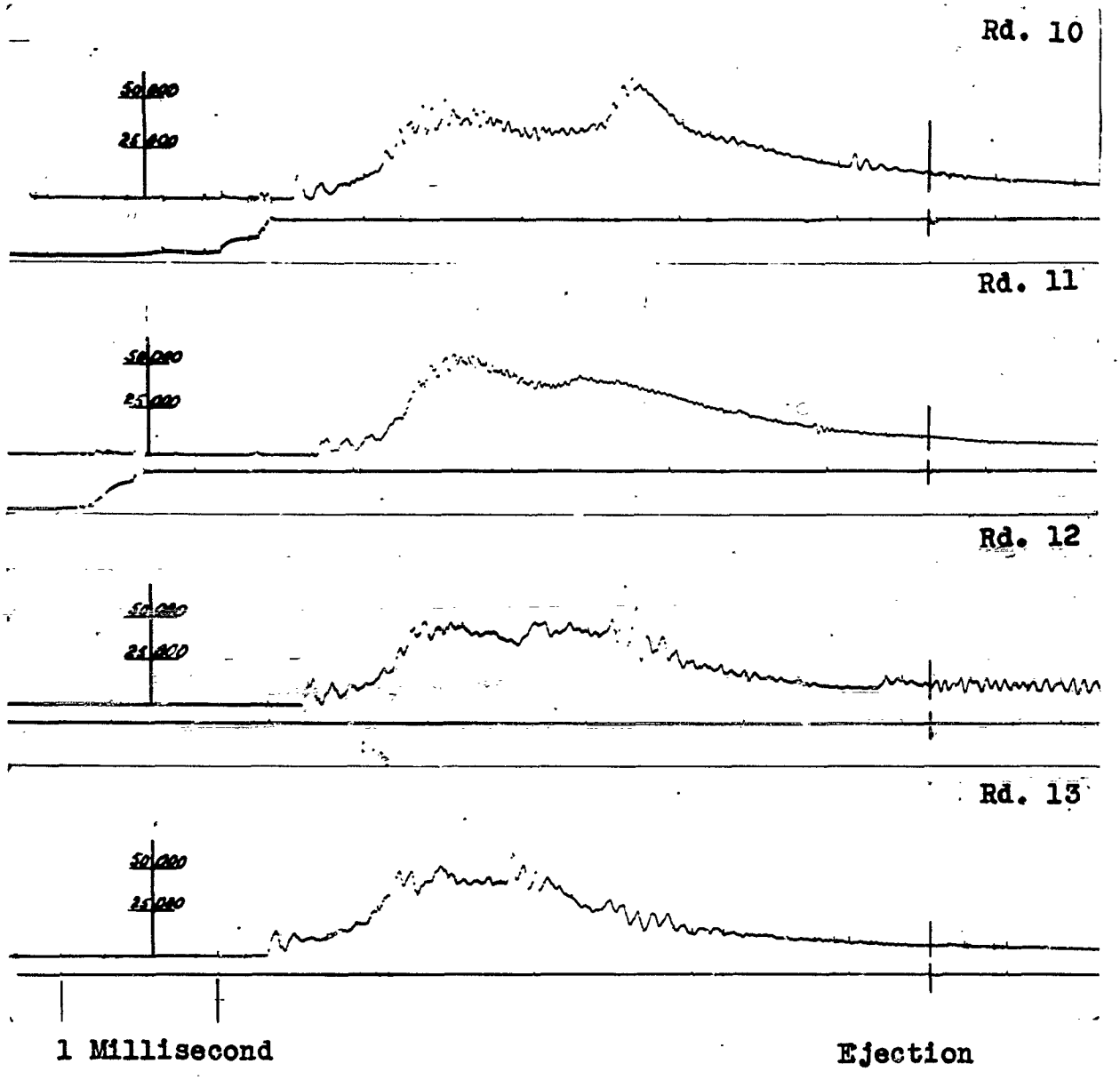


FIGURE NO. 6

PRESSURE-TIME OSCILLOGRAMS OBTAINED WITH ASSEMBLY C-5
AT THE 3" GAGE POSITION

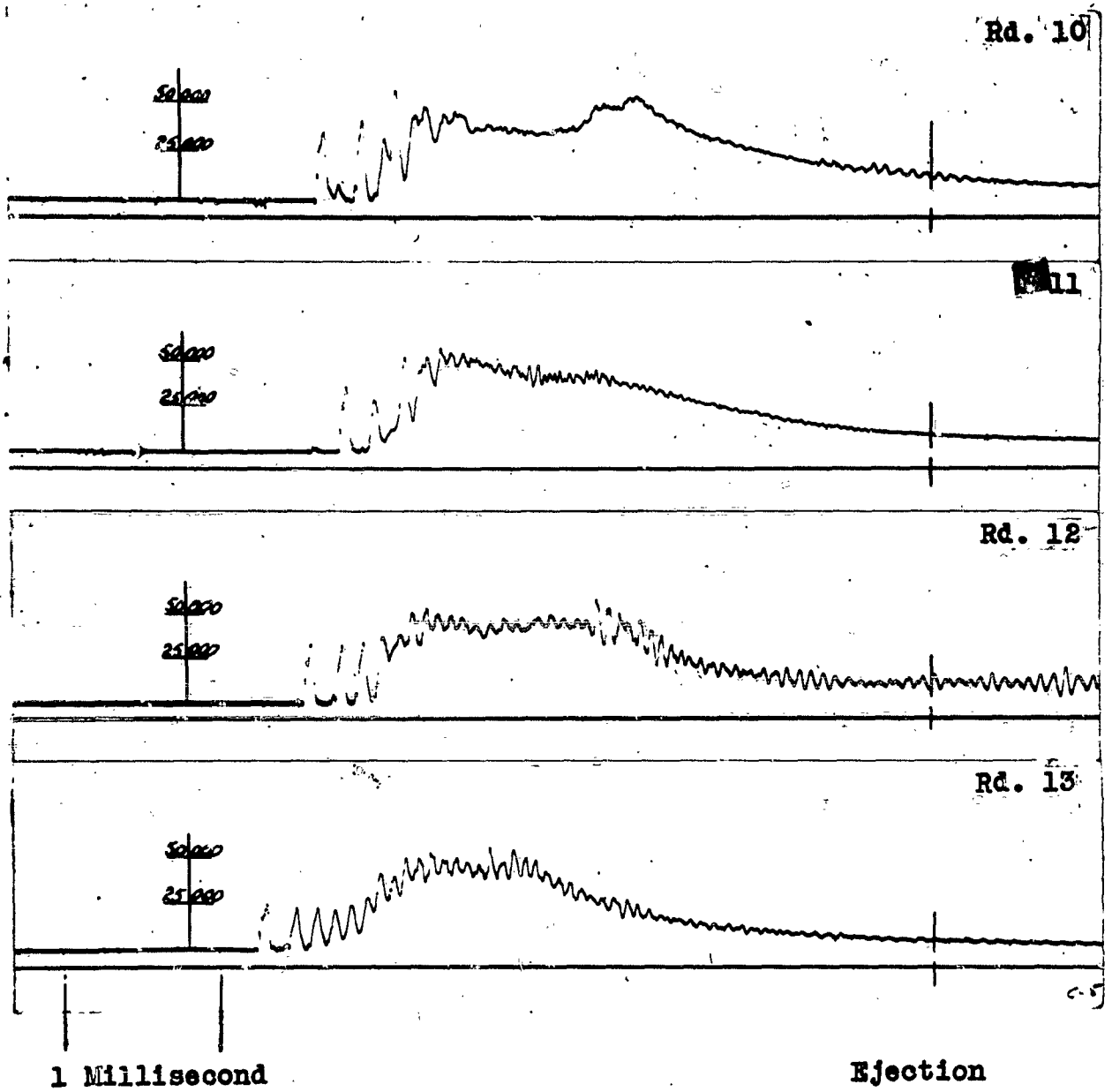


FIGURE NO. 7

PRESSURE-TIME OSCILLOGRAMS OBTAINED WITH ASSEMBLY C-5
AT THE 10" GAGE POSITION

The C-5 Assembly

The general tendency with the C-5 case assembly, as with the C-4, was for the second pressure peak to diminish in amplitude relative to the first as the point of venting was advanced in the propellant column. However, the drop in pressure between the two maxima is relatively much less in this series and close approximations to plateau conditions were obtained on rounds 6, 7, 11 and 12.

A marked decrease in muzzle velocity resulted in this series as the venting point of the primer was advanced in the case. From an average of 3020 f/s on the two rounds with 4"0 primers the velocity dropped to 2173 f/s on the round with the 8"0 primer.

TABLE 2 BALLISTIC DATA FROM C-5 ASSEMBLY

Rd. No.	Primer Length	Ejection Time (ms)	Pressure (psi) First Peak 3"0 Gauge	First Peak 10"0 Gauge	Second Peak 3"0 Gauge	Second Peak 10"0 Gauge	Velocity (f/s)
5	4"0	4.0	36850	34200	49100	39750	3038
6	5"0	4.0	46050	36900	41450	35050	2852
7	6"0	4.1	48600	52450	43050	---	2750
8	7"0	4.1	44600	40200	42250	---	2389
9	8"0	4.4	43500	30200	28100	16850	2173
10	4"0	4.0	43500	39600	54550	48685	3001
11	5"0	3.9	51100	51800	41650	42395	2867
12	6"0	4.0	42150	43300	45520	42550	2752
13	7"0	4.2	45200	47050	43300	46650	2484

General

In comparing the results from the two case assemblies, it is to be noted that the second pressure peak can be eliminated and an essentially plateau type pressure curve produced by proper location of the venting point of the primer in the propellant column. However, the amount of unburned propellant remaining in the case is increased as venting is advanced. This condition would be particularly undesirable under rapid fire conditions. In the C-5 assembly, advancing the point of venting results in a marked

reduction in the velocity obtained with this assembly. Thus, while advancing the point at which the primer vents in the propellant column improves the ballistics of the round in some respects, it has detrimental effects in others. The C-5 assembly with venting of the primer near the base of the propellant column is the most satisfactory of the rounds tested with respect to velocity obtained, shape of pressure curves, and amount of propellant utilized.

CONCLUSIONS

From the firings of these tests, it is concluded that advancing the point of venting of the primer within the propellant column produces the following effects:

a. Muzzle velocity is increased in rounds with a propellant column L/D ratio of 5.0, but is decreased in rounds with a ratio of 3.4.

b. The amount of unburned propellant remaining in the case after firing increases in both case configurations. Negligible amounts of propellant were observed in cases of either configuration in which the primers vented near the base of the propellant column.

c. In both case configurations, the pressure curves were transformed from curves with two distinct pressure peaks, through stages approximating plateau configurations, to curves with one well defined pressure maximum.

REFERENCES

(a) BUORD ltr NP9-Re5a-FWB:fl of 15 Jul 1952
(b) BUORD Conf ltr NP9-Re2d-WES:aph Ser 49271 of 17 Dec 1952
(c) NPG Conf Report No. 1200 of 13 Nov 1953

APPENDIX A

DISTRIBUTION

Bureau of Ordnance:

Ad3	1
Ad8	4
Re2	1
Re2d	2
Re5	1
Re5e	2

Inspector of Naval Material
Development Contract Department
Redwood City, California 2

Superintendent
U. S. Naval Gun Factory
(Aircraft Armament Section)
Washington 25, D. C. 1

Director of Jet Propulsion Laboratory
California Institute of Technology
Pasadena 3, California
Via: Inspector of Naval Material
1206 South Santee Street
Los Angeles 16, California 1

Redel Incorporated
7405 Varna Street
North Hollywood, California
Via: Inspector of Naval Material
1206 South Santee Street
Los Angeles 16, California 1

Commander
U. S. Naval Ordnance Laboratory
White Oak, Silver Spring, Maryland 1

DISTRIBUTION (Continued)

Mathieson Chemical Corporation
Research Division
Niagara Falls, New York
Via: Inspector of Naval Material
Hotel Buffalo, 14th Floor
Washington and Swan Streets
Buffalo 3, New York 1

Experiment Incorporated
Richmond, Virginia
Via: Naval Inspector of Ordnance
Johns Hopkins University
8621 Georgia Avenue
Silver Spring, Maryland 1

Detroit Controls Corporation
Research Division
806 Chestnut St.
Redwood City, California 1

Chief of Ordnance (ORDTA)
Department of the Army
Washington 25, D. C. 2

The Armour Research Foundation
Technology Center
Chicago 16, Illinois
Via: Inspector of Naval Material
205 W. Monroe Street
Chicago 6, Illinois 1

Bureau of Ordnance Technical
Liaison Officer
Southern California Area
1030 East Green Street
Pasadena 1, California 1

Bureau of Ordnance Technical
Liaison Officer
Aberdeen Proving Ground
Aberdeen, Maryland 1

DISTRIBUTION (Continued)

Office of Naval Research
(Armaments Branch)
Department of the Navy
Washington 25, D. C. 1

Bureau of Aeronautics
Armament Division (AR-70)
Department of the Navy
Washington 25, D. C. 1

Chief of Ordnance (ORDTU)
Department of the Army
Washington 25, D. C. 1

Chief of Ordnance (ORDTS)
Department of the Army
Washington 25, D. C. 1

Armed Services Technical Information Agency
Document Service Center
Knott Building
Dayton 2, Ohio 5

Commanding Officer
Frankford Arsenal
(Pittman-Dunn Laboratory)
Philadelphia, Pennsylvania
Attn: M. W. Silverstein 1

Commanding General
Aberdeen Proving Ground
Aberdeen, Maryland
Attn: Technical Information Section
Development and Proof Services 1

Commanding General
Wright Air Development Center
Wright-Patterson Air Force Base
(WCLG) Armament Laboratory
Dayton, Ohio 1

DISTRIBUTION (Continued)

Cornell Aeronautical Laboratories Inc.
4455 Genesee Street
Buffalo 21, New York
Via: Bureau of Aeronautics Representative
Cornell Aeronautical Laboratory
Box 235, Buffalo 21, New York 1

Commanding Officer (R and D(PC))
Springfield Armory
Springfield, Massachusetts 1

Commanding General
Air Research and Development Command
Directorate of Armament (RDDR)
Baltimore, Maryland 1

Commanding Officer
U. S. Naval Air Rocket Test Station
Lake Denmark
Dover, New Jersey 1

Purdue Research Foundation
Lafayette, Indiana
Attn: Professor J. T. Agnew
Via: Inspector of Naval Material
3802 So. Calhoun Street
Fort Wayne 6, Indiana 1

Olin Industries
Winchester Division
New Haven, Connecticut
Attn: R. S. Holmes
Via: Inspector of Naval Material
Building 23C
1285 Boston Avenue
Bridgeport 8, Connecticut 1

DISTRIBUTION (Continued)

Olin Industries
Western Cartridge Division
East Alton, Illinois
Attn: D. S. Ryker
Via: Inspector of Naval Material
Room 701 Buder Building
707 Market Street
St. Louis 1, Missouri 1

Catholic University of America
7th and Michigan Avenue, N. E.
Washington 17, D. C.
Attn: Dr. V. Griffing
Via: Inspector of Naval Material
401 Water Street
Baltimore, Maryland 1

Chief of Ordnance (ORDTR)
Department of the Army
Washington 25, D. C. 1

Chief of Ordnance
Department of the Army
Washington 25, D. C.
Attn: ORDTT 2

Local:

OMI	1
OMIR	2
OML	1
OR	1
OM-1	1
File	1

Subject: Ignition of Liquid Propellant in the 40mm Gun
by G. L. Poudrier and K. H. Crutchfield, Armament Department, U. S. Naval Proving Ground, Dahlgren Virginia
15 June 1954

ABSTRACT

The tests reported here were conducted to determine the effects produced on the ballistics of a liquid monopropellant round by advancing the point of venting of the primer within the propellant column. The tests were conducted with two case configurations in which the L/D ratios of the propellant columns were approximately 3.4 and 5.0. As the point of venting was advanced, muzzle velocity increased in rounds with the L/D ratio of 5.0, but decreased in those with the L/D ratio of 3.4. The amount of unburned propellant remaining in the case after firing increased in both case configurations. In both configurations, the pressure curves were transformed from curves with two distinct pressure peaks, through stages approximating plateau configurations, to curves with one well defined pressure maximum.